

IN THE CLAIMS

Please amend claim 15 as follows:

1-11. (Canceled)

12. (Currently Amended) A gas turbine combustor comprising:

a combustion chamber supplied with fuel and air, the fuel and air being supplied to said combustion chamber as a plurality of coaxial flows; and

a plurality of modules, each being composed of a plurality of fuel nozzles and air holes corresponding to said plurality of fuel nozzles, fuel jet holes of said fuel nozzles being arranged coaxially or approximately coaxially with said corresponding air holes, respectively, and each of said fuel jet holes and each of said corresponding air holes being formed so that a fuel jet flow from each of said fuel jet holes is jetted toward a center or a position close to the center of an inlet center of each of said corresponding air holes and the fuel jet flow and an annular air flow encompassing the fuel jet flow are jetted into said combustion chamber.

13. (Previously Presented) A gas turbine combustor according to claim 12, wherein said plurality of modules are fluidly separated from each other and fluidly connected to fuel supply systems, respectively, and a control unit for controlling individually said fuel supply systems according to gas turbine load is provided.

14. (Previously Presented) A gas turbine combustor according to claim 12, wherein a swirling angle which provides a swirling component around the axis of the combustor is given to a part of or all of said fuel nozzles and corresponding air holes.

15. (Currently Amended) A gas turbine combustor comprising:

a combustion chamber; and

a plurality of modules, each being composed of a plurality of fuel nozzles each for jetting fuel into said combustion chamber and air holes each for jetting air into said combustion chamber, said air holes being arranged in a combustion chamber wall surface, and said fuel nozzles and said air holes being arranged so that the fuel and air are

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- jetted into said combustion chamber as a plurality of coaxial flows.